

TO: EPO, The Netherlands  
DATE: April 7, 2005  
RE: International Patent Application PCT/DE2004/001137  
Title: Rotor for a Gas Turbine and Gas Turbine  
Our File: P803328/WO/1

In response to the Written Opinion of the International Search Authority (Rule 43 bis. 1 PCT) dated October 5, 2004:

**BOX No. II - Priority** of the Written Opinion discusses the fact, that a copy of the earlier application upon which priority is claimed, has not yet been submitted.

In an effort to remove this objection, a copy of the earlier application is enclosed with the present, written response.

**BOX No. V – Reasoned Statement...** of the written opinion affirms novelty and inventive steps with respect to claims 7 and 8. As such, please accept herewith for filing a new scope of patent protection to now include five (5) claims. The subject matter of new claim 1 with certain clarifications now combines the features found in original claims 1 through 4. The characterizing portion of new claim 1 is identical as concerns contents to the characterizing portion of the original claim 7. Dependent claims 2 through 5 which substantially correspond in contents to the original claims 8 through 11 now read upon new claim 1. We deleted the features of original claims 5, 6, and 12.

All amendments are shown in detail in the enclosed version (Red Pen Correction) showing the amendments with hand-written notations.

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/s/  
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Enclosures

Patent Claims 1 through 5, page 7 (single copy), via facsimile and regular mail  
Red Pen Correction, 2 pages (single copy), via facsimile and regular mail  
Reproduction of the earlier Application 12 pages (single copy), via regular mail

CLAIMS:

1. Rotor for a gas turbine, having several blades (12a, 13a, ... 29b, 30b) arranged on the rotor (10) and rotating together with the rotor (10), that is, having running blades, the blades forming a blade ring, the blades (12a, 13a, ... 29b, 30b) within the blade ring being arranged at a different distance from one another and thus with a different blade pitch, the distance between the blades (12a, 13a, ... 29b, 30b) within the blade ring changing continuously or discontinuously in the circumferential direction, and the distance between the blades (12a, 13a, ... 29b, 30b) within the blade ring being dimensioned such that imbalances cancel one another out, characterized in that the rotor has several blade rings arranged axially behind one another, and in that, within each blade ring, the blades are arranged at a different distance from one another.
2. Rotor according to Claim 1, characterized in that the arrangement of blades within the respective blade ring is different for each blade ring.
3. Rotor according to one or more of Claims 1 or 2, characterized in that the rotor is constructed as a turbine rotor

or compressor rotor of a gas turbine, particularly of an airplane engine.

4. Rotor according to one or more of Claims 1 or 2, characterized in that the rotor is constructed as a fan rotor of a gas turbine, particularly of an airplane engine.

5. Rotor according to one of the preceding claims, characterized in that the rotor is constructed as a blisk (bladed disk) or bling (bladed ring) of a gas turbine, particularly of an airplane engine, the blades being an integral component of the rotor.